

# ASK 21



Flight Manual  
Instructions For Continued  
Airworthiness  
Repair Manual



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## FLIGHT MANUAL

# SCHLEICHER ASK 21

This Manual must be carried on board at all times.

Registration : **N 8 9 9 S A**  
.....

Factory serial number : **2 1 9 4 6**  
.....

Owner : .....  
.....  
.....  
.....

This Flight Manual is FAA approved for U.S. registered gliders in accordance with the provisions of 14 CFR Section 21.29 and is required by FAA Type Certificate Data Sheet No. G 47 EU 1.10.83  
.....

German edition of this Manual is approved under § 12(1)2 LuftGerPO.

Published **March 9, 1983**

Approval of translation has been done by best knowledge and judgement. In any case the original text in German language is authoritative.



*Grüß*  
- 9/ März 1983

# ASK 21 Flight Manual (US-version)

## Record of Revisions

Rev No.	Pages Affected	Date of Approval	Date of Insertion	Ref. / Signature
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TN 11	2, 14, 15, 42, 43	20.12.83	19.03.84	Juw
TN 13	13	16.02.84	28.02.84	Juw
TN 13a	2, 13	04.06.84	18.06.84	Juw
TN 14	2, 21	16.05.84	08.06.84	Juw
TN 15	2, 25, 26, 26a, 27	25.05.84	10.06.84	Juw
TN 20	2, 3, 4, 30, 31, 42, 42a, 43	16.10.87	06.11.87	Juw
TN 21	annex (new tow release coupling)	17.01.90	18.03.90	Juw
TN 22	2, 42a, 43a, 43b	04.12.90	20.12.90	Juw
TN 23	2, 22, 24, 33, 34	29.01.91	18.02.91	Juw
AD 93-001/2/3	annex (L'Hotellier instructions for the maintenance)	27.04.98	15.05.98	Juw
TN 25	2, 25, 30	16.02.93	28.06.93	Juw
TN 28	annex (nose bolt)	14.03.00	16.04.00	Juw



# ASK 21 Flight Manual (US-version)

TN 32	17c	01.06.10	23.08.10	mg
TN 38	25, 26a, 27a, 27b, 27.1a, 30	31.05.16	10.2017	



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(Beginning of JAR22-required and LBA-approved part.)

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#### Appendix

Flight polar

Lubrication Scheme

Rigging data (for adjustment of control surfaces, etc.)

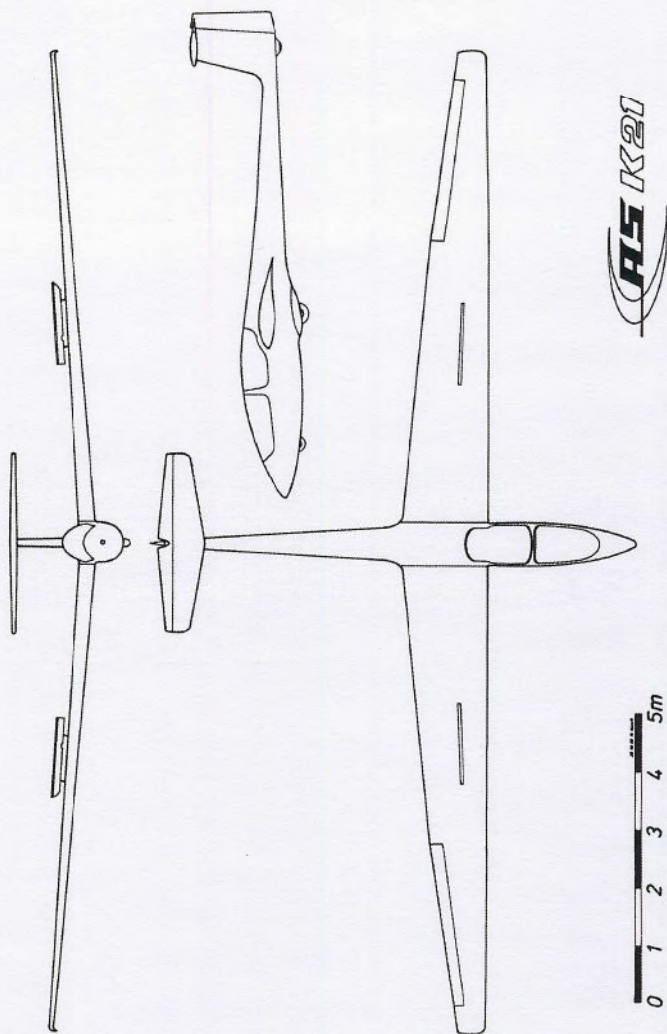
Spin training with the ASK 21 – Summary of important information  
and further recommendations

Excerpt of the USAF Manual



I.4

THREE-SIDE-VIEW



## I.5 DESCRIPTION

The ASK 21 is designed to meet the needs of modern gliding training. It has an all fiberglass sandwich structure.

Midwing with T-tail, tandem seat arrangement, airbrakes on upper wing only.

The glider is stressed for aerobatics (inverted flight included).

Technical Data

Span	17,00 m	=	55,74 ft
Length	8,35 m	=	27,4 ft
Height	1,53 m	=	5,02 ft
Aspect ratio	16,1		
Wing area	17,95 m <sup>2</sup>	=	192,96 sqft
Max. all up weight	600 daN	=	1320 lbs
Max. wing loading	33,4 daN/m <sup>2</sup>	=	6,84 lbs/sqft

Airfoil: Wortmann FX 302 196 (inner wing)  
Wortmann FX 60 -126 (wing tip)

Winch Tow: Weak Link 1000 daN

Aero Tow : Weak Link 600 daN

## **II. OPERATING LIMITATIONS**

### **II.1 Airworthiness Category**

A (Aerobatics) according to LFSM.

Certification basis: Airworthiness Requirements for Sailplanes and Powered Sailplanes dated 1.11.1975.

### **II.2 Permitted Operations**

The glider is certified for VFR flights during daytime (VFR day).

The approved operation class is indicated by a data placard on the instrument panel. Depending on the respective equipment the glider may be licensed for traffic for the following categories:

1. Airworthiness Category U (Utility), according to VFR with equipment as under 11.3 a)
2. Airworthiness Category A (aerobatics), with equipment as under 11.3 a) and 11.3 b) for the following aerobatics:

Loop, Stall Turn, Split 'S',  
Immelmann, Slow Roll, Inverted Flights,  
Spin, Steep Climbing Turn, Lazy Eight,  
Chandelle.

With spin ballast attached, aerobatics are prohibited (except spinning).



II.3 MINIMUM EQUIPMENT

- a) 2 airspeed indicators;  
2 altimeters;  
2 four-point safety harnesses;  
2 seat cushions, at least 10 cm thick when loaded,  
or parachutes (automatic or manual);  
Weight & balance data placard for both seats;  
Data plate;  
Flight Manual.
  
- b) Additional equipment for aerobatics  
Bottom straps for safety harnesses in both seats;  
1 G-meter for front seat;  
Foot loops on rudder pedals;  
Parachute (automatic or manual).

## II.4 AIRSPEED LIMITATIONS AND LOAD FACTOR LIMITS

Max. permissible speed (calm air):

$$V_{NE} = 151,2 \text{ kts} = 174,00 \text{ mph} = 280 \text{ km/h}$$

Max. permissible speed (rough air):

$$V_B = 108,0 \text{ kts} = 124,3 \text{ mph} = 200 \text{ km/h}$$

Max. maneuvering speed:

$$V_M = 97,2 \text{ kts} = 112,0 \text{ mph} = 180 \text{ km/h}$$

Max. speed with airbrakes extended:

$$V_{LE} = 151,2 \text{ kts} = 174,00 \text{ mph} = 280 \text{ km/h}$$

Stall speed with airbrakes extended:

$$V_{S1} = 37,0 \text{ kts} = 42,3 \text{ mph} = 68 \text{ km/h}$$

Stall speed with airbrakes retracted:

$$V_{S0} = 35,0 \text{ kts} = 40,4 \text{ mph} = 65 \text{ km/h}$$

The following safe load factors must not be exceeded (airbrakes retracted, symmetrical maneuvers):

$$\text{At max. maneuvering speed } V_M \quad n = \begin{matrix} +6,5 \\ -4,0 \end{matrix}$$

$$\text{At max. permissible speed } V_{NE} \quad n = \begin{matrix} +5,3 \\ -3,0 \end{matrix}$$

Rough air is defined as turbulence that can be expected in wave rotors, thunderstorms, whirlwinds, and when crossing mountain ridges.

Maneuvering speed is the highest speed at which full deflections of the control surfaces are still permitted.

With max. permissible speed  $V_{NE}$  only 1/3 of the possible deflections are permitted.

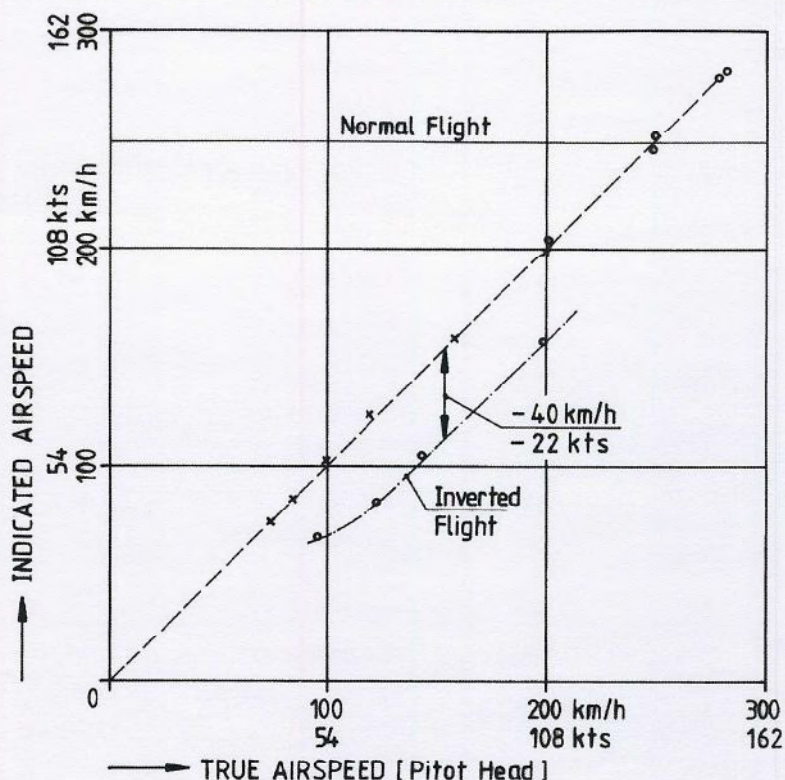
True airspeed (TAS) is, however, relevant for safety against flutter. Therefore, one must take into account that with increasing altitude the true airspeed is higher than the reading of the airspeed indicator because of the decreasing air density.

$$V_{NE} = 151 \text{ kts} \quad n = \begin{matrix} +5,3 \\ -3,0 \end{matrix}$$

$V_{NE}$  at various altitudes

Altitude ft	$V_{NE}$	
	knots	mph
5000	151	174
10000	144	165
15000	132	152
20000	121	139





#### POSITION ERROR

With normal flights the position error of the airspeed indicator is negligible within the whole range up to 280 km/h ( 151 kts ).

With inverted flights the airspeed indicator reads too low, i.e. up to -40 km/h ( 22 kts ).

By attaching an extension tube this error may be eliminated. (see also pages 27/28).

The extension tube must project at least 70 mm ( 2,75 in ) past the fuselage nose.

Airspeed indicator markings (IAS)

Red line (max. permissible airspeed):

151,2 kts = 174,0 mph = 280 km/h

Yellow arc (caution range):

97,2 - 151 kts = 112 - 174 mph = 180 - 280 km/h

Green arc (normal range):

43,0 - 97 kts = 50 - 112 mph = 80 - 180 km/h

Yellow triangle (approach speed):

49,0 kts = 56,0 mph = 90 km/h

II.5 CREW : 2 persons

Minimum crew : 1 person (min.weight 70 daN = 154 lbs)

Caution: Solo flights may only be conducted from  
the front seat !

II.6 WEIGHTS

Empty weight approx. 792 lbs = 360 daN

Max. all up weight 1320 lbs = 600 daN

Max. weight of non lift  
producing members 902 lbs = 410 daN.

## II.7 IN FLIGHT CENTER OF GRAVITY RANGE

The approved in flight C.G. range is from  
 9,21 (234 mm) - 18,46 inches (469 mm) behind the  
 datum line; equivalent to 20 % - 41,1 % of the MAC =  
 44,13 inches (1121 mm). With a 0,31 inches (8 mm)  
 behind leading edge center part of the wing.

## II.8 WEIGHT &amp; BALANCE INFORMATION

Max. payload front seat (pilot incl. parachute):  
 242 lbs = 110 daN.

Min. payload front seat (pilot incl. parachute):  
 154 lbs = 70 daN.

Caution: Short weight in the front seat must be  
 compensated by ballast (installation of lead discs  
 in the nose; 1 lead disc = 2,76 lbs pilot weight).

Number of lead discs	Min. payload front seat	
	daN $\approx$ kg	lbs
0	70,0	154,32
1	68,75	151,57
2	67,5	148,81
3	66,25	146,06
4	65,0	143,30
5	63,75	140,54
6	62,5	137,79
7	61,25	135,03
8	60,0	132,28
9	58,75	129,52
10	57,5	126,77
11	56,25	124,01
12	55,0	121,25

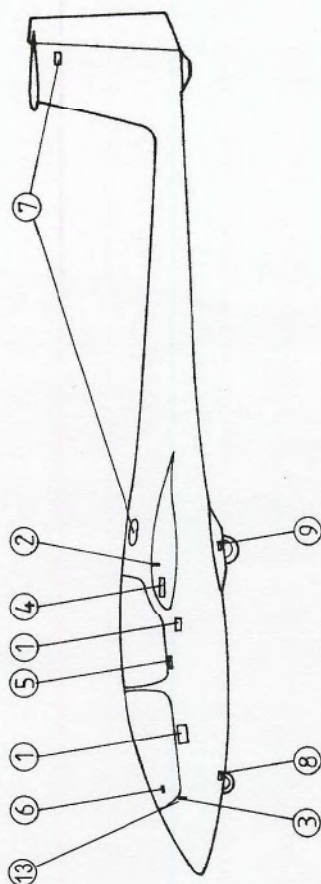
Max. payload rear seat (pilot incl. parachute) :  
 242 lbs = 110 daN.

1 kg = 2,2046223 lbs

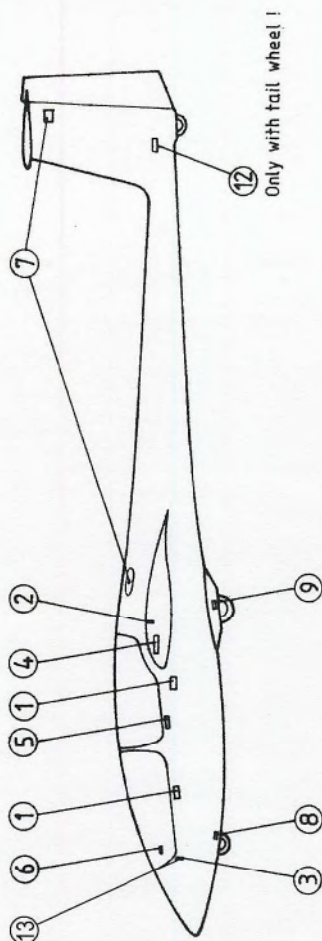


# Settings of placards

## II.9



## Settings of placards [Only with tail wheel]



**Segelflugzeugbau A. Schleicher Poppenhausen**

Model \_\_\_\_\_ Serial no. \_\_\_\_\_

**DATA PLACARD**

Approved for:

Max. speed for calm air	280 km/h
Max. speed for rough air	200 km/h
Max. maneuvering speed	$V_M$ 180 km/h
Max. aero tow speed	$V_F$ 180 km/h
Max. winch launch speed	$V_W$ 150 km/h

**WEIGHT AND BALANCE**

Min. payload front seat	kg
Max. payload front seat	kg
Max. payload rear seat	kg
Baggage in wingroots	max. 2 x 10 kg
Max. permissible all-up weight	kg

2 off

Loading of baggage compartment: max. 10 kg

2 off

**Pre Take Off Check:**

1. Tail dolly removed – ballast checked?
2. Parachute properly fastened – raise line?
3. Safety harness properly fastened – all operating elements within reach?
4. Put your toes under the toe-straps! Do not flatten the straps! Danger of jamming the pedals!
5. Airbrakes retracted and locked?
6. Placard for spin ballast?
7. Altimeter adjusted?
8. Radio on – frequency and volume checked?
9. Trim adjusted?
10. Control circuit check – Controls easy to operate?
11. Airspace for start and release clear?
12. Check wind
13. Prepared for take-off interruption?
14. Both canopies closed and locked? –  
Emergency jettisoning procedure in mind?

1 off



⑤

1 off  
Rear

**Attention! Emergency bailout!**

- a) Pull back both canopy side-locks and push canopy upwards.
- b) Undo safety harness.
- c) Get up and bail out.
- d) With manual chute seize release grip and pull out entirely after 1-3 sec.

②



**A. Schleicher**  
6416 Poppenhausen

Model : ASK 21  
Serial no: 21 XXX  
Registration letters :  
Made in West Germany

③

1 off

**Aerobatics prohibited!**  
Equipment as under airworthiness category "U" (Utility)

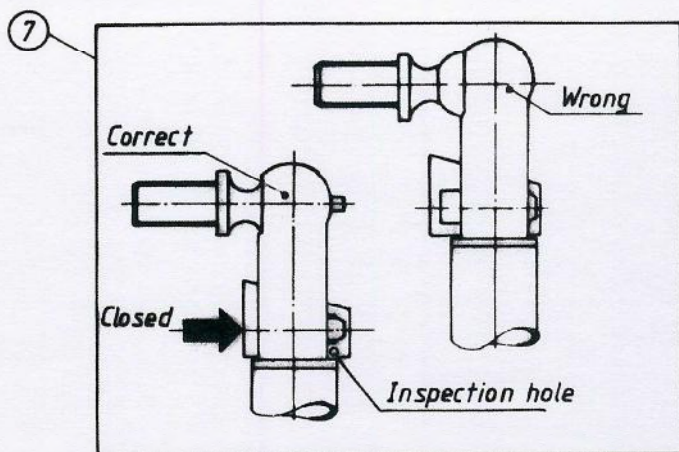
For equipment without  
g-meter and bottom strap.

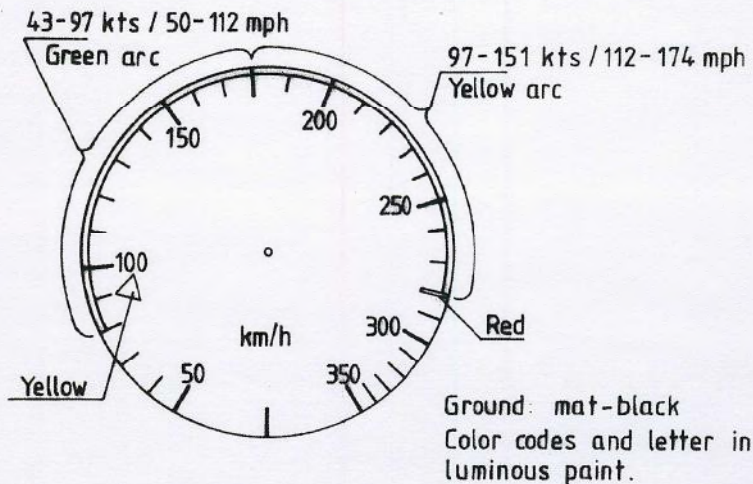
③

1 off

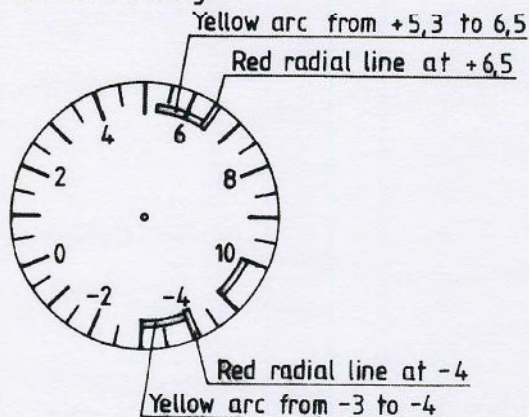
**Aerobatics as per Flight Manual**  
Equipment as under airworthiness category "A" (Acrobatic)

For equipment with  
g-meter and bottom strap.



Airspeed indicator color codesG-meter color codes

## a) Positive range



## b) Negative range



## II.11 DESCRIPTION OF SYMBOLIC PLACARDS



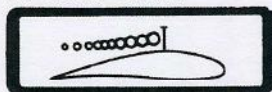
Rudder pedals adjustment: grey knob on RH side of the console.

To adjust pedals backwards:

Take your feet off the pedals and pull pedals backwards; then let go the grey knob and load the pedals in order to lock them.

To adjust pedals forwards:

Pull grey knob and push pedals forwards with your heels; then let go the grey knob and load the pedals in order to lock them.



Airbrakes: blue lever in the LH arm rest; pull to extend airbrakes.



Trim: noseheavy.



Trim: tailheavy.



Tow release: yellow knob LH below canopy frame.



To open canopy: pull back the white levers LH and RH on the canopy frame.





Canopy emergency jettisoning:  
push to the left the red flat knob  
above the instrument panel



Ventilation

Prior to take off, check  
proper engagement of  
the canopy locks!

This placard must be fitted in the  
front and rear cockpit in full view of  
the pilot.

When the plane is equipped with an attachment for spin ballast  
(TN4b)

Placard at the front instrument panel, informing about mounted spin  
ballast.



A M8-screw must be  
mounted through the plac-  
ard from the back. The  
placard is visible, when  
spin ballast is mounted on  
the tail (= DANGER).  
When spin ballast is re-  
moved, the placard is cov-  
ered by the nut that other-

wise holds the spin ballast.



### III. EMERGENCY PROCEDURES

#### III.1 RECOVERY FROM SPIN

According to the standard procedure spinning is terminated as follows:

- a) Apply opposite rudder (i.e. apply rudder against the direction of rotation of the spin).
- b) Short pause (hold control inputs for about 1/2 spin turn).

**Warning:** Disregarding the pause will result in slower recovery!

- c) Ease the control column forward (i.e. give in to the pressure of the stick) until the rotation ceases and sound airflow is established again.

**Warning:** Full forward stick may retard or even prevent the recovery!

- d) Centralise rudder and allow glider to dive out.

The altitude loss from the beginning of the recovery until the normal flight attitude is regained is about 80 meter (260 feet).

**Note:** During spins the ASK 21 oscillates in pitch. From a steep nose down spin recovery according to the standard procedure is up to 1 turn, from a flat spin less than 1 turn.

#### III.2 CANOPY JETTISONING AND EMERGENCY BAIL OUT

##### Front canopy:

- a) Move lever with the red knob above the instrument panel to the left and push canopy upwards.
- b) Open safety harness.
- c) Get up and bail out.
- d) With manual chute seize release grip and pull out entirely after 1 to 3 seconds.

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Author    Date  
Kaiser    Mar. 9. 1983

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Rear canopy

- a) Pull back both canopy side locks and push canopy upwards.
- b) Open safety harness.
- c) Get up and bail out.
- d) With manual chute seize release grip and pull out entirely after 1-3 seconds.

If circumstances allow, the front pilot should allow the rear pilot to bail out first.

### III.3 FLIGHTS THROUGH PRECIPITATION

With wet or slightly iced wings or with insect accumulation there will be no deterioration in flight characteristics.

However, one has to reckon with a rather considerable deterioration in flight performance. This must be taken into account especially on landing final approach.

Add a safety margin of 5 knots = 10 km/h for approach speed !

## III.4 WING DROPPING

The glider stalls extremely benign. Nevertheless one always has to face the possibility of wing dropping because of turbulence. In that case push stick forward immediately and apply opposite rudder against a noticeable turn at the same time to regain a normal flight attitude. If the rudder deflection against the turn is forgotten, a spin may occur even if the stick pressure is released.

## III.5 GROUND LOOPING

For normal conditions, smooth runway, short grass, one may take off with the wing on the ground without having to fear a change in the direction.

High grass and rough ground, however, may cause ground looping. In that case release the tow rope immediately.



#### IV. NORMAL OPERATING PROCEDURES

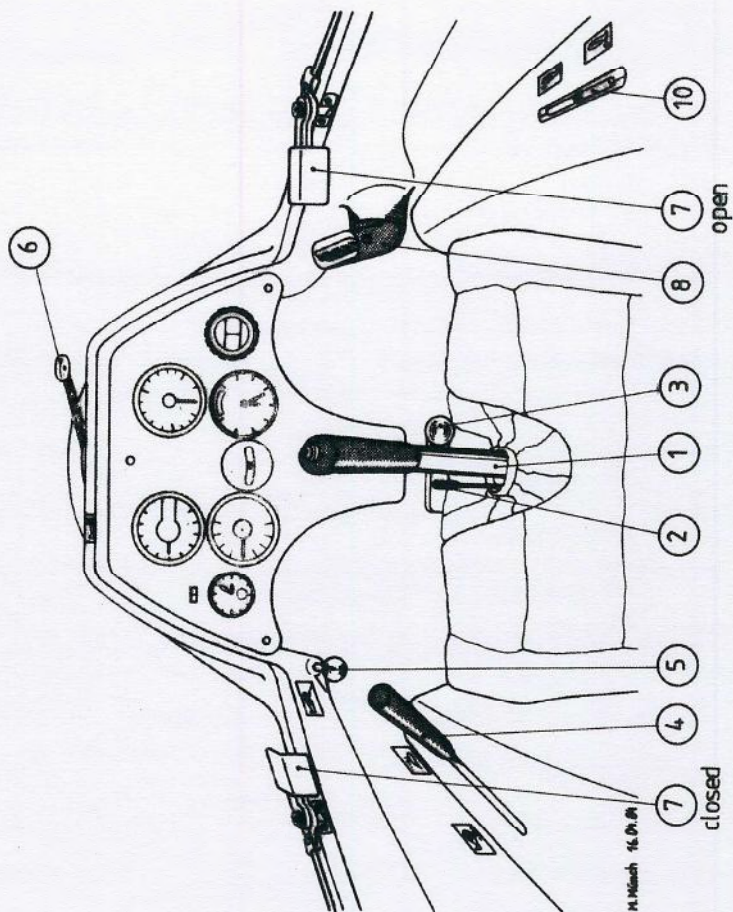
##### IV.1. COCKPIT LAYOUT AND CONTROLS

###### Front seat:

- No.1: Stick.
- No.2: Trim; flat lever with **green** knob LH of stick.
- No.3: Rudder pedal adjustment; **grey** knob at the console.
- No.4: Airbrakes with wheel brake; **blue** lever in the left lever arm rest.
- No.5: Release cable; **yellow** knob on left cockpit wall below the canopy frame.
- No.6: Canopy emergency jettisoning; horizontal lever with **red** flat grip above the instrument panel cover; to the left = OPEN.
- No.7: **Front** canopy locking:  
**White** swivel levers on left and right canopy frame.  
To open canopy: pull **back** both levers.  
To lock canopy: push both levers **forwards**, parallel to the canopy frame.
- No.8: Ventilation nozzle; on right cockpit wall below the canopy frame; revolving and lockable.
- No.9: Back rest; the back rest is adjustable by tilting it from the bottom upwards and forwards (see sketch); in normal flight attitudes the backrest cannot shift by itself.  
Very tall pilots may fly without the back rest.
- No.10: Trim indicator; in the right arm rest behind the ventilation nozzle.
- No.11: Detachable rudder hand lever at the left cockpit wall below the airbrake grip (not shown). Only applicable for mod. TN No. 25 and TN No. 30.



Front seat



Rear seat:

- No.1: Stick.
- No.2: Trim; flat lever with **green** knob LH of stick.
- No.3: Rudder pedal adjustment with circular grip in front of stick.
- No.4: Airbrakes with wheel brake; **blue** lever in the left lever arm rest.
- No.5: Release cable; **yellow** knob on left cockpit wall below the canopy frame.
- No.6: Rear canopy locking = Canopy **emergency jettisoning**; **red** swivel levers on left and right canopy frame.  
To open canopy: pull **back** both levers.  
To lock canopy: push both levers forwards, parallel to the canopy frame
- No.7: Ventilation nozzle; on right cockpit wall below the canopy frame; revolving and lockable.
- No.8: Back rest; the back rest is adjustable by tilting it from the bottom upwards and forwards (see sketch); in normal flight attitudes the backrest cannot shift by itself.  
Very tall pilots may fly without the back rest.
- No.9: Trim indicator; in the right arm rest behind the ventilation nozzle.
- No.10: Detachable rudder hand lever at the left cockpit wall below the airbrake grip (not shown). Only applicable for mod. TN No. 30.

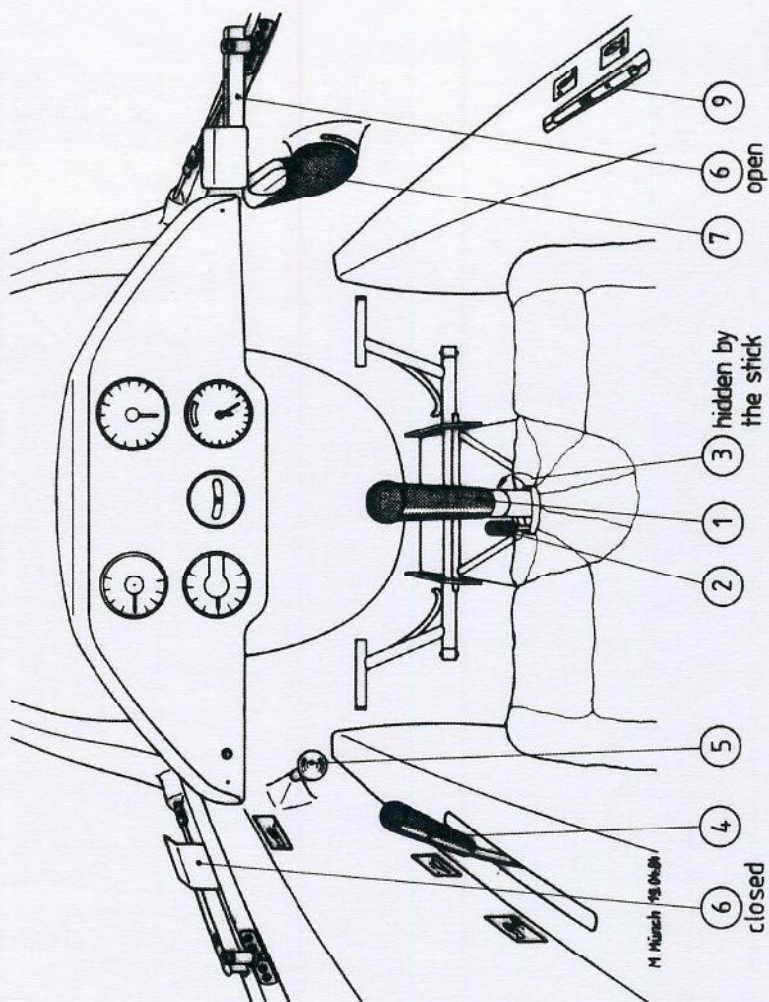
Adjustable Headrest:

On the front seat there is an adjustable headrest. When the backrest is used, the headrest is inserted into the base part at the backrest. The headrest is secured in the correct height with the locking pin. Without backrest, the headrest is inserted through the cutout in the instrument cover of the rear instrument panel. A ball catch must lock in noticeable.

The headrest is adjusted correctly, when the head can touch the headrest on eyelevel.



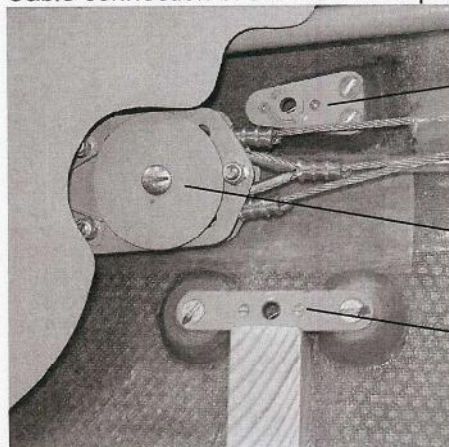
Rear seat



Connection of the rudder pedals:

The rudder control with a hand lever enables pilots to fly the ASK 21, even when their capability to move their legs is limited. In this case the feet resting on the pedals must not obstruct the rudder. Therefore, the cables coming from the particular pedals can be disconnected at the joint between the front and rear seat. The disconnected cables have to be fixed on special anchor points on the fuselage wall.

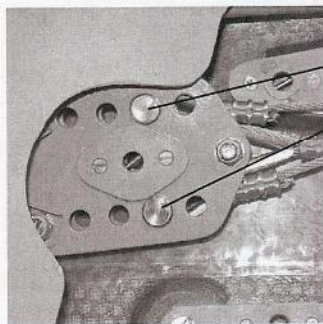
Cable connection of the left side is pictured:



Anchor point for disconnected cable of the front pedal.

Connection plate for cables of rudder pedals with lock washer for connection bolts.

Anchor point for disconnected cable of the rear pedal.

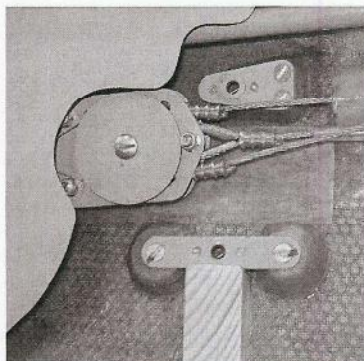


Connection plate for cables of rudder pedals with demounted lock washer. The removable connection bolts are visible.

**Attention:** The position of the connection bolts in the row of holes must be marked, to avoid a change of the total cable length when reconnecting, which would result in a change of the pedal position. The holes not needed will already be closed ex works.

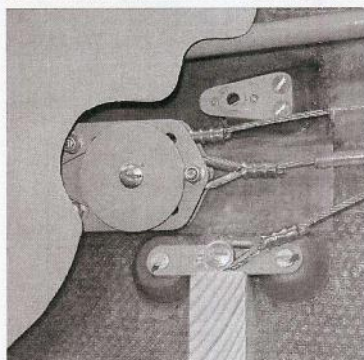


## Possible Variants:

**Front and rear pedals connected****Attention:**

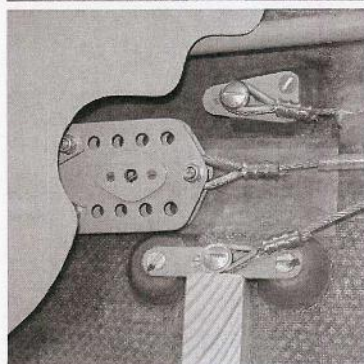
The lock washer must be mounted according to the rules to hold the connection bolts in their position.

The screws of the anchor points for the disconnected cables must be removed!

**Rear pedals disconnected****Attention:**

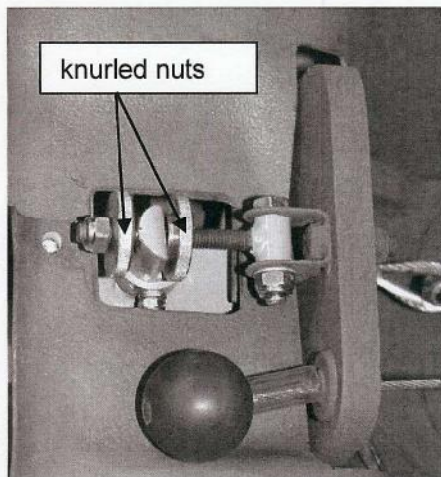
The towing ropes of the **rear** pedals have to be attached with screws left and right at the **lower** anchor points.

Likewise also only the **front** pedals can be disconnected and have to be fixed on the **upper** anchor points.

**Front and rear pedals disconnected**

In this situation, the rudder can in both pilot seats only be controlled with a hand lever.

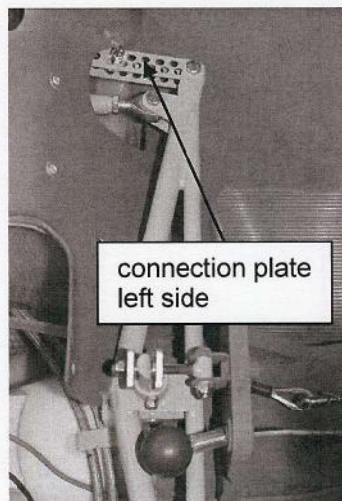


Checking and Adjusting of the Cable Tension:

The adjusting screw for the cable tension of the rudder hand control system is located in the front of the control column and besides of the pedal adjustment. This is for the **fine adjustment** of the cable tension during the Pre Flight Check.

Shifting in the direction of flight results in a higher cable tension, shifting against a lower. After the adjustment both knurled nuts have to lock the part between again.

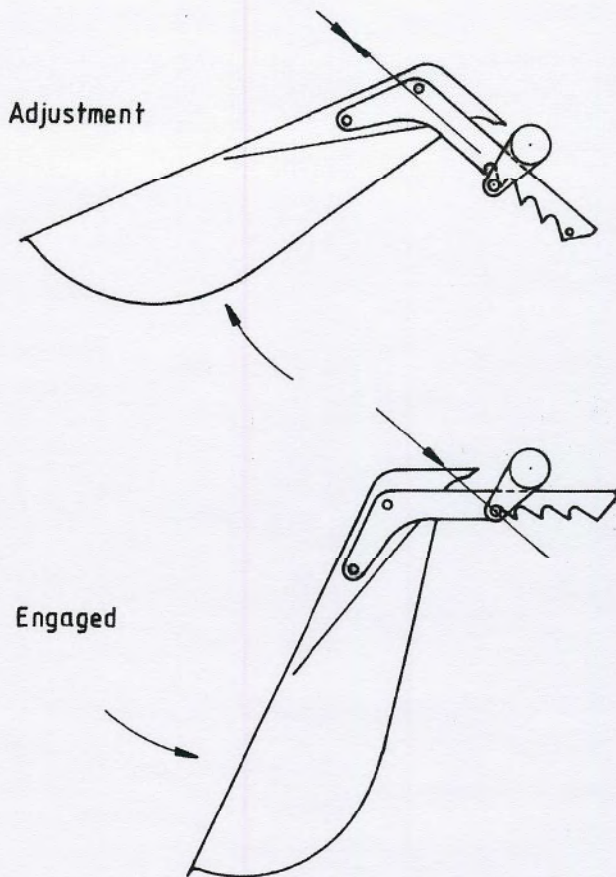
The cable tension is high enough, if the hand lever has no noticeable play with blocked rudder. The cable tension is too high, if the rudder does not move freely.



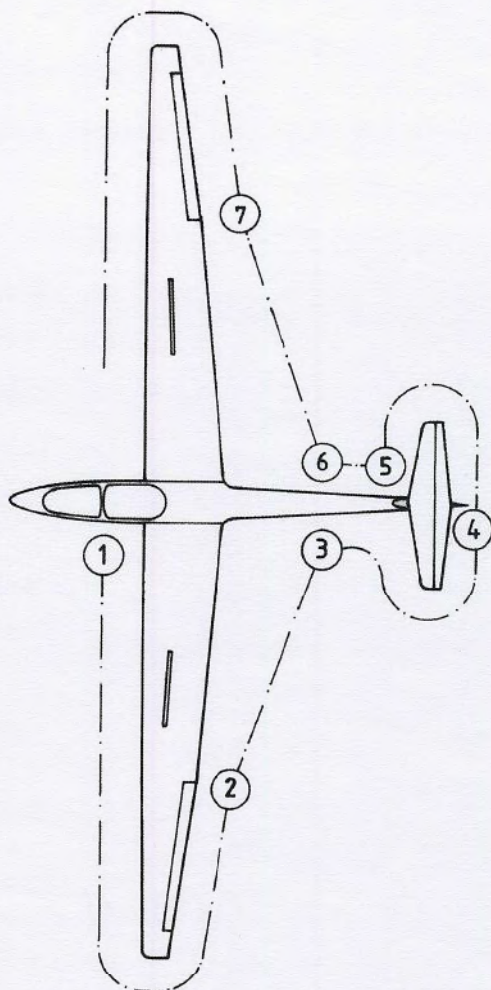
If the adjusting screw allows no correct adjustment of the cable tension, the system has to be maintained. It has to be adjusted and inspected as follows:

- Remove the front fairings covering the swing lever.
- Adjust the cable tension by readjusting the cable mounting points on connection plates left and right on the swing lever.
- Check freedom from play and free movement.

### BACK REST ADJUSTMENT



DAILY CHECKS [see page 30 FM ]





## IV.2 DAILY INSPECTIONS

Prior to flight operations the following checks must be accomplished:

- 1.a. Open canopy! Check that the main pins are properly secured.
- b. Check the proper connection of the ailerons and airbrakes through the access hole on the left side above the wing. Are the quick-release connectors secured with spring clips?
- c. Check for foreign bodies!
- d. Check the control circuits force and that all controls are free-moving. Apply full deflections and load the control circuits with fixed controls and air-brakes. Check the plastic tubes inside the S-shaped rudder pedal tubes for proper and tight fit.
- e. Check tire pressure:
  - Nose wheel 2.0 bar (28psi)
  - Main wheel 2.7 bar (38psi)
  - Tail wheel (If installed) 2.5 bar (35.6 psi)
- f. The condition and function of the tow release mechanism is to be checked. Actuate the tow release: does it snap back freely? Engage and disengage the ring pair. Check the automatic release of the C.G. towing hook with the ring pair which must release automatically backwards.
- g. Check the wheel brake. Pull the air brake lever; at the end of its travel an elastic resistance must be felt.
- h. Check cable tension of the rudder hand control system (see page 27.1a).
  - Only in flight operation with the rudder hand lever:
    - Locking bolt screwed in at the airbrake handle and secured?
    - Rudder hand lever mounted and secured?
    - Are the rudder cables connected to the joint between both seats and secured resp. disconnected and attached to the anchor points? (only TN 30)
  - Only in flight operation without rudder hand lever:
    - Are the rudder cables connected to the joint between both seats and secured according to the rules? (only TN 30)
- 2.a. Check upper and lower wing surface for damages!
- b. Aileron: its condition, free-movingness and play is to be checked! Check also the pushrod connection.
- c. Airbrake: its condition, fit and locking is to be checked.
3. Check the fuselage for damages, in particular also the bottom side.
4. Check that the tailplane is properly assembled and secured. Check also the pushrod connection! Secured with spring clips?

5. Check condition of tailskid, pitot tube and venturi tube.
6. Check static vents for cleanness !
7. After rough landings or excessive flight stress the whole sailplane must be checked with the wings and the tail unit being removed (see also point 2.)!). If any damage is found, a technical aviation inspector must be called in. On no account one must take off again before such damage has been repaired.  
See also the Instructions For Continued Airworthiness !!

#### IV.3. CHECKS PRIOR TO TAKE OFF

See the Check Lists in Section VII., p.43, of the Instructions For Continued Airworthiness !



*[Handwritten signature]*  
- 3. NOV, 1987



## IV.4 TAKE OFF

Winch tow

Trim neutral.

Max. tow speed: 81 kts = 93 mph = 150 km/h.

The glider features a tow release for winch tow in front of the main wheel.

The most favorable tow speed is 50-60 kts = 56-58 mph = 90-110 km/h.

There is little pitch up tendency during initial tow. In the upper third of the tow additional altitude may be gained by slight back pressure.

Tow release: pull the release knob several times to the stop.

Aero tow

Aero tows only at the nose release in front of the nose wheel. Recommended tow rope length: 100-200 ft.

Trim neutral.

Max. tow speed: 97 kts = 112 mph = 180 km/h.

The most favorable tow speed during climb is 50-75 kts = 56-87 mph = 90-140 km/h.

Take off may be done with the wingtip on the ground. Getting the wings level is no problem. However, the pilot is advised to be careful with high grass and very rough ground.

Lift off takes place at about 40 kts = 47 mph = 75 km/h.



## IV.5 FREE FLIGHT

The glider may be flown up to  $V_{NE} = 151 \text{ kts} = 174 \text{ mph} = 280 \text{ km/h}$ .  
Up to manoeuvring speed of  $97 \text{ kts} = 112 \text{ mph} = 180 \text{ km/h}$  full control deflections can be applied. At higher speeds the controls must be applied more carefully.  
At  $V_{NE}$  only 1/3 of the max. possible deflections must be applied.

## IV.6 LOW SPEED FLIGHT, WING DROPPING AND SPINS

With the stick back a distinct tail buffet is felt.  
The glider is very benign in low speed flight. By use of normal aileron deflections the wings may be kept level down to minimum speed, even with aft C.G. positions.  
With normal rudder deflections no wing dropping is found. Yaw angles of up to  $5^\circ$  have no significant influence on the wing dropping attitude.

Also rapid pulling up into  $30^\circ$  pitch does not cause wing dropping, but only a gentle nose drop. The same applies for stalling out of a  $45^\circ$  turn.

But one has to point out that even the most benign glider needs speed in order to be controllable.

In turbulence this is especially important when also a wing dropping may occur.

Spin development from wing dropping strongly depends on the C.G. position and also to some extent from the pilot reaction.

For C.G. positions forward of 315 mm aft of datum the ASK 21 does not spin at all. This configuration applies to 2 heavy pilots.

For C.G. positions from 320 mm through 385 mm aft of datum more incipient spin turns are possible followed by self recovery after 4 1/2 turns at most. Such C.G. positions are possible in dual flight with a lightweight pilot in the front seat.

For C.G. positions aft of 400 mm behind datum controllable sustained spins are possible. Such a C.G. position is usually only possible with one lightweight pilot in the front seat.

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Author    Date  
Kaiser    Mar. 9. 1983

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**NOTE:** During spins the ASK 21 oscillates in pitch. From a steep nose down spin recovery according to the standard procedure is up to 1 turn, from a flat spin less than 1 turn.

The speed at which the stall takes place depends on the payload. The following standard values are applicable:

	without airbrakes	with airbrakes
Single, all up weight 1043 lbs = 470 daN	65 km/h 35 kts IAS	68 km/h 37 kts IAS
Dual, all up weight 1320lbs = 600daN	74 km/h 40 kts IAS	77 km/h 42 kts IAS

#### Spinning with spin ballast

Mounting of spin ballast see pages 47a and 47b. Spinning with spin ballast is only allowed by dual flights. With spin ballast other aerobatic manoeuvres are not permissible.

#### *Entry procedure:*

The best entry speed is 2 km/h (1.1 kts) above the speed, at which the stall warning sets in. This must be checked before in flight.

Step hard on the rudder in the intended spin direction. Then, fully pull the stick. The aileron stays neutral. The rudder must stay in this position as long as the spin is supposed to continue.

**WARNING:** *If a spiral dive sets in, it must be stopped immediately, to prevent overstressing the structure.*

#### *Recovery procedure:*

Recovery according to the standard procedure, see chapter III.1.

Further information can be found in the appendix to the Flight Manual, titled "Spin training with the ASK 21: Summary of important information and further recommendations".



#### IV.7 High Speed Flight

The sailplane shows no flutter tendency within the permissible speed range.

With airbrakes extended in a 45° dive the speed remains below  $V_{NE} = 280 \text{ km/h}$  (151 kts); it goes up to 232 km/h (125 kts) at  $G = 600 \text{ kg}$  (1323 lbs).

#### IV.8 Approach and Landing

The most favorable approach speed is 49 kts = 56 mph = 90 km/h. With turbulence it may be advisable to increase slightly the approach speed.

Even steep approaches may be slowed down efficiently with the airbrakes at the beginning of the landing final approach.

NOTE: The airbrakes increase the stalling speed by about 1,6 kts = 3km/h.

Sideslipping is also suitable as an approach control. With full rudder during the sideslip the rudder pressure decreases to zero; the rudder must be pushed back.

During sideslip the airspeed indication goes to zero reading.



## IV.9 AEROBATICS

Warning : Even a glider which is approved for full aerobatics does not have infinite strength capacities. Most hazardous are aerobatics which get out of control or are badly executed, as they result in high loads.

Therefore, it is urgently recommended to have oneself guided by an experienced flight instructor. The ASK 21 being an approved two-seater for full aerobatics offers this possibility.

Such guidance is even prescribed according to § 69 (4) of the German LuftPersPO (Aviation Personnel Test Regulations) dated January 9, 1976. Following § 96 (3) of the said LuftPersPO an adequate experience is required from flight instructors.

Note : the normal airspeed indicator system shows a large pressure error in inverted flight during which the airspeed indicator reads 40 km/h = 22 kts too low. When extending the pitot head by attaching a brass tube - 12 Ø x 1; 5,5 in = 140 mm in length - this error disappears. The tube must project in the front at least 2,75 in = 70 mm. For normal flights this is not necessary. In order to avoid damage when parking the glider in the hangar, this tube should not be left on any longer than necessary.

Permissible indicated speeds

Inverted flight without pitot head extension:

V <sub>NE</sub> :	Single	35-130kts = 65-240km/h.
	Dual	38-130kts = 70-240km/h.
Indicated maneuvering speed		75kts = 140km/h
Indicated max. speed		130kts = 240km/h.

Inverted flight with pitot head extension:

Indicated maneuvering speed		97kts = 180km/h
Indicated max. speed		151kts = 280km/h
Indicated stall speed		47kts = 87km/h
With two occupants		

**ATTENTION:** *never release stick and rudder pedals when flying aerobatics.*

For aerobatics instruction a reliable agreement must be made between instructor and student flyer with regard to the communication system for the mutual taking over of the controls.

Airbrakes must be extended as soon as the pilot loses the control of the glider or as the speed increases involuntarily too fast.

Exception: "Tail sliding"!!!

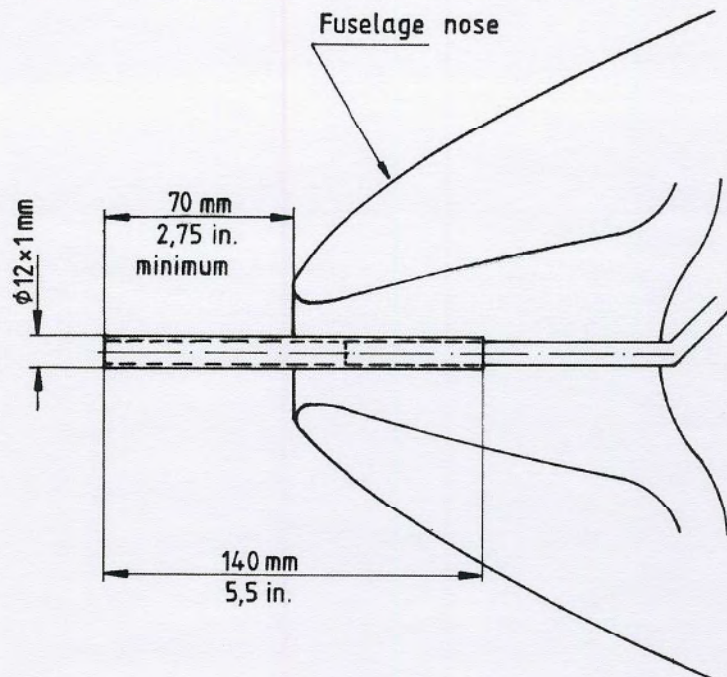
The trim remains in the center position for aerobatic manoeuvres. Do not ever change the trim when flying aerobatics!!

With spin ballast is attached, aerobatics are prohibited (except spinning).

**PROHIBITED AEROBATICS:**

All abrupt aerobatic maneuvers  
Loop forward  
Tail sliding.

Extension tube for total pressure head with inverted flights



Brass tube 5,5 in = 140 mm in length (12  $\phi$  x 1).  
One may also use a suitable plastic tube provided  
that it is sufficiently stiff and straight.



<u>Recommended entrance speeds for the following aerobatics</u>		<u>Max.acceleration</u>
Loop upward	Single: 84 kts = 155 km/h	2-3 g
	Dual: 92 kts = 170 km/h	
Stall Turn	Single: 89 kts = 165 km/h	3 g
	Dual: 97 kts = 180 km/h	
Split 'S'	Single: 92 kts = 170 km/h	2-3 g
	Dual: 97 kts = 180 km/h	
Immelmann	Single: 89 kts = 165 km/h	2, 5-3, 5 g
	Dual: 97 kts = 180 KM/h	
Slow Roll	Single: 81 kts = 150 km/h	
	Dual: 89 kts = 165 km/h	
Steep Climbing Turns & Lazy Eight	Single: 76 kts = 140 km/h	
	Dual: 81 kts = 150 km/h	
Chandelle	Single: 86 kts = 160 km/h	
	Dual: 95 kts = 175 km/h	



### L O O P

Entrance speed:

Single 84 kts = 155 km/h

Dual 92 kts = 170 km/h

Max. g = 2-3.



### S T A L L T U R N

Entrance speed:

Single 89 kts = 165 km/h

Dual 97 kts = 180 km/h

Max. g = 3.



### S P L I T 'S'

Pull up at least 30°!

Altitude loss approx. 328 ft = 100 m.

Entrance speed:

Single 92 kts = 170 km/h

Dual 97 kts = 180 km/h

Max. g = 2-3.



### I M M E L M A N N

Entrance speed:

Single 89 kts = 165 km/h

Dual 97 kts = 180 km/h

Max. g = 2,5-3,5.



S L O W   R O L L

Entrance speed:

Single 81 kts = 150 km/h

Dual 89 kts = 165 km/h.



I N V E R T E D   F L I G H T

Note: with the inverted flight the fuselage nose will be unexpectedly high above the horizon.



S P I N



L A Z Y   E I G H T

Entrance speed:

Single 76 kts = 140 km/h

Dual 81 kts = 150 km/h.





S T E E P   C L I M B I N G  
T U R N

Entrance speed:

Single 76 kts = 140 km/h

Dual    81 kts = 150 km/h.



C H A N D E L L E

Entrance speed:

Single 86 kts = 160 km/h

Dual    95 kts = 175 km/h.

## V. RIGGING AND DE-RIGGING

## V.1 RIGGING

Rigging the ASK 21 can be carried out by four persons without mechanical assistance, and by three persons with the use of a fuselage stand or a wing support.

Prior to rigging, clean and grease all pins, bolts, bushings and control system connections.

1. Set up the fuselage and hold it horizontal.
2. Plug the spar fork of the left wing into the fuselage and - if available - place a wing support under the wing end.
3. Offer up the right wing and align the main pin fittings.
4. Press in the main pins and secure. Never insert the rear wing attachment pins prior to the main pins !
5. Press in the rear wing attachment pins; unscrew the T-tool and check whether the safety lock is engaged.
6. Connect and lock the aileron control linkages in the fuselage behind the spar tunnel. You must be able to touch the ball pivot by feeling through the slot in the socket. Also check the proper engagement of the safety lock by pushing it on to close ! Secure them with spring clips!
7. Connect and lock the airbrake control linkages in the fuselage behind the spar tunnel. Secure them with spring clips!



8. The tailplane is fitted onto the fin from the front. (see Fig. V.2-1 and V.2-2).

Now the Allan bolt at the leading edge is screwed in; this should be screwed in tightly until the spring-loaded safety pin snaps out over the screw head as far as the socket.

9. Connect the elevator and safety with a spring clip !

**Note,** if your glider uses an automatic elevator connection: after cleaning and lightly greasing the plug-in elevator connections, the tailplane is fitted onto the fin from the front; both elevator panels must be fitted into their connectors simultaneously. Then the tailplane is pushed back until the Allan bolt at the leading edge can be screwed in; this should be screwed in tightly until the spring-loaded safety pin snaps out over the screw head as far as the socket.

10. Carry out a pre-flight check referring to the Check List (see Section VII, p.43, of the Instructions For Continued Airworthiness !

11. The control circuits must be subjected to an operational test.

12. Check condition and function of the wheel brake; check the tire pressure.

See also Section IV.2 DAILY INSPECTIONS in this Manual.



## V.2 DE-RIGGING

De-rigging is carried out in the reverse sequence to that of rigging. It must be taken care that the rear wing attachment pins have to be removed prior to the main pins.

**WARNING:** For derigging the horizontal tail from the fin it has to be regarded that only the method according to Fig. V.2-2 is used.

Fig. V.2-1  
WRONG: Twist movement

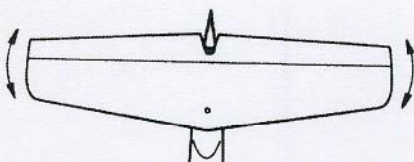
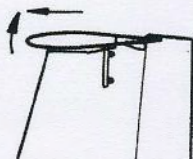


Fig. V.2-2  
RIGHT: Pitch movement



## V.3 PARKING

When parking the glider, the canopies have to be closed !

When an ASK 21 is parked on an airfield in the sunshine (this must also be observed during the waiting time until take-off when the pilots are already on board) the canopies must not be left open for some time.

Depending on the position of the sun and the intensity of the radiation, the burning-glass effect of the canopies can cause a slow fire in the area of the instrument panel or the headrest respectively.

Therefore, if you have to store the glider outside, it is absolutely necessary always to close the canopies and to cover them with a white cloth.

#### V.4 ROAD TRANSPORT

The design of a glider trailer is another subject and cannot be discussed in all details here. Of course, a closed trailer is preferable. But also an open trailer may serve the purpose, the latter is generally simpler and lighter. It is important that all components are well fixed and have a large support surface.

A structural components survey drawing which can be used for the building of a trailer, can be obtained from ALEXANDER SCHLEICHER.

**WARNING:** In no case must the elevator actuator fitting be loaded. This fitting trades out of the upper end of the fin. Not even soft foam cushions are allowed.

For the construction of the trailer for road transport the full freedom from any load must be carefully regarded.

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## V.5 PREVENTIVE MAINTENANCE

The whole surface of the glider is painted with a weather resisting, white polyester coat. Impurities may be washed off with a mild cleansing agent. Heavy impurities may be removed with a polish. For the paint maintenance only silicone-free agents must be used (e.g. 1 Z-special cleansing agent-D2 from W.SAUER & CO., 5060 Bensberg, West Germany, - or the cleansing polish from LESONAL).

Though the glider is rather insensitive, it should be protected as much as possible against moisture and humidity. If water has soaked into any components, these have to be stored in a dry room and must be turned over frequently;

The canopy is best cleaned with a special plexiglass cleansing agent; in an emergency lukewarm water will do. Rewipe only with pure, soft leather or with glove cloth. Never wipe on dry plexiglass.

The safety harnesses must be regularly checked for damage and tears. The metal parts of the harnesses must be checked for corrosion.



## VI. CENTER OF GRAVITY ( CG )

## VI.1 WEIGHING PROCEDURE OF CG AT EMTY WEIGHT

Prior to determining the CG in flight the CG at empty weight has to be established by weighing the glider. For this procedure the glider must be put on two pair of scales (one at the nose wheel and one at the tail skid).

NOTE: the glider must be set on the two pairs of scales very carefully in order to prevent that the scales get misaligned; (this could lead to erroneous results).

The Datum Line (DL) is situated at the wing leading edge of the straight center part of the wing.

Levelling means: wedge on rear top edge of fuselage 1000 : 52 horizontal.

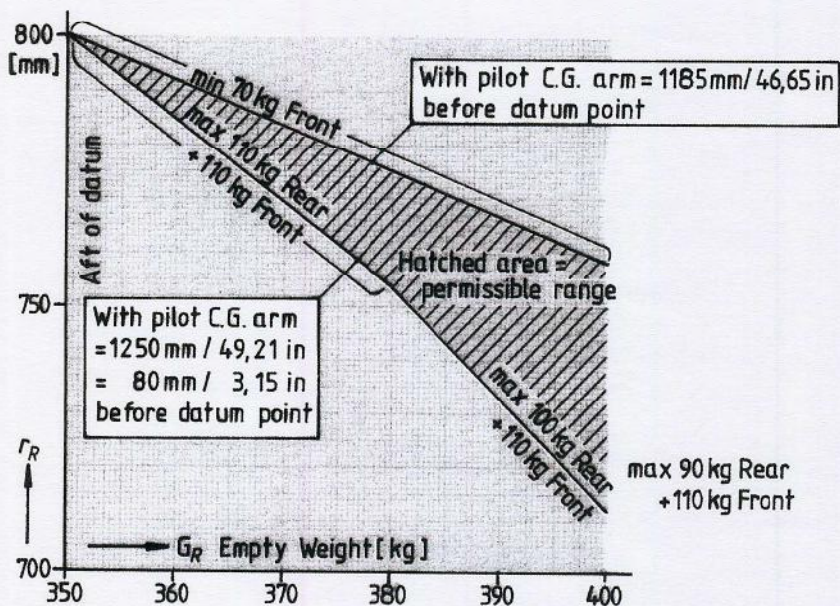
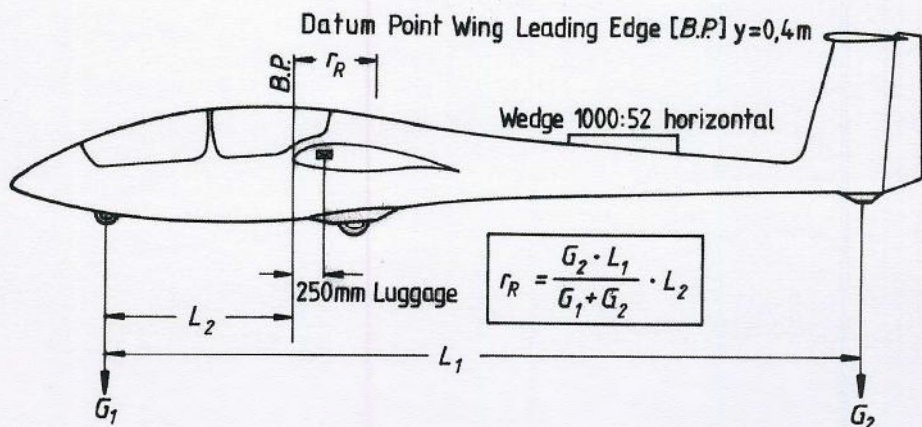
Empty weight CG :

Weight at the nose wheel:	lbs
Weight at the tailskid:	lbs
Support point nose wheel:	in
Support point tailskid:	in

NOTE: determination of empty weight and empty weight CG must be done without any additional balance weights (e.g. trim cushion).

Be careful not to exceed the maximum weight of non lift producing parts when using maximum payload. The total weight of non lift producing parts contains the individual weights of fuselage, elevator and maximum payload and must not exceed 410 daN = 920 lbs (the payload must be reduced accordingly).

Weight and Balance Sheet





The CG should be recalculated after repair, repainting or installation of additional equipment, but not later than 4 years after the last weighing.

The empty weight, empty weight CG position and maximum load should be recorded after each weighing on page of the Flight Manual by a competent person.

## VI.2 EMPTY WEIGHT CG POSITION

With the empty weight CG according to the below-mentioned limits and the pilot weights according to the load table, the in flight CG will be within the approved range.

Empty Weight		CG forward		CG aft	
daN	lbs	mm	in	mm	in
350	770	800	31,50	800	31,50
360	792	784	30,87	792	31,18
370	814	769	30,28	783	30,83
380	836	754	29,69	774	30,47
390	858	732	28,82	766	30,16
400	880	712	28,03	758	29,84



Weight and balance information with spin ballast

Without valid spin ballast-table (Flight Manual page 47b), spin ballast at the tail may not be used. The validity period is specified on each spin ballast-table. A valid spin ballast table can be obtained from the manufacturer (procedure, refer to Maintenance Manual page 36).

Before every flight with spin ballast the pilots must be weighed with the equipment worn in flight (clothes, parachute ...).

When the load in the front seat is below 70kg (154lbs), compensate missing load by attaching trim ballast in the front fitting, so that the load in the front seat equals 70kg (154lbs). For this purpose, follow the instructions on page 13. During the further procedure, the front pilot and the front trim ballast count together as a pilot of 70kg (154lbs).

The amount of spin ballast is specified in the current spin ballast-table. The mass of the pilot in the front seat defines the line of the table; the mass of the pilot in the rear seat defines the column of the table. At the intersection, the number of ballast plates (1 kg = 2.2 lbs), which are to be attached, is noted.

Up to a maximum of 12 spin ballast plates are permissible. The plates have to be distributed evenly to the left and right side of the fin and have to be fixed with the provided screw.

**CAUTION:** *The washer and nut fixed at the cockpit placard (see below) **must** be used. After removal of the spin ballast the washer and the nut must be fixed again on the placard.*

**Attention****Check spin ballast!**

Only use spin ballast for flights  
with two pilots!

This table gives the number of weights to be mounted at the tail for spin training (section IV.6)

Serial No: 21946

Empty mass: 853 lbs

Max permissible payload: 470 lbs

Registration: N899SA

Empty mass Centre of Gravity: 29.2 inch

Date: 26.10.17

Weigh and Balance report dated: 19.10.17

Equipment list dated: 19.10.17

pilot mass incl. parachute in rear seat (lbs)										pilot mass incl. parachute in front seat (lbs)										
154	159	163	168	172	176	181	185	190	194	198	203	207	212	216	220	225	229	234	238	243
154	4	4	4	4	4	5	5	5	5	6	6	6	6	6	7	7	7	7	7	8
159	4	5	5	5	5	5	6	6	6	6	6	7	7	7	7	7	8	8	8	8
163	5	5	5	6	6	6	6	7	7	7	7	7	8	8	8	8	8	9	9	9
168	6	6	6	6	7	7	7	7	7	8	8	8	8	8	9	9	9	9	9	10
172	6	6	7	7	7	7	8	8	8	8	8	9	9	9	9	9	10	10	10	10
176	7	7	8	8	8	8	9	9	9	9	9	9	10	10	10	10	10	11	11	11
181	8	8	8	8	9	9	9	9	10	10	10	10	10	10	11	11	11	11	11	12
185	8	8	9	9	9	9	10	10	10	10	10	11	11	11	11	11	12	12	12	12
190	9	9	9	10	10	10	10	11	11	11	11	11	11	12	12	12	12	12	12	12
194	10	10	10	10	11	11	11	11	11	12	12	12	12	12	12	12	12	12	12	12
198	10	10	11	11	11	11	12	12	12	12	12	12	12	12	12	12	12	12	12	12
203	11	11	11	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
207	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
212	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
216	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
220	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
225	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
229	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
234	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
238	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12
243	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12	12

= invalid loading

Note: This table becomes invalid when the aircraft is modified or with the date of: 19.10.2021

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TN 4b 12.Dec.12 JN/MG

drawn up by:



## VI.3 Weighing Record


Date of weighing, carried out by	10.10.2017	Equipment list used for weighing (date)	10.10.2017	Empty weight (kg) (lbs)	387 (853,2)	Empty CG behind datum mm (in)	741 (29,17)	Empty weight momentum max	286,739	Max pay-load kg (lbs) max	213 (470)	Old spin ballast table* removed (check off)	N/A	Signature	RAC see next pg
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\* Flight Manual Page 47b

The empty weight momentum is necessary to calculate the in flight cg. (load table).



# ASK 21 Flughandbuch

Datum d. Wägung 10.10.2017	Leergewichts-Schwerpunkt hinter BP in mm 741	Vord. Sitz Zuladung incl. Fallschirm in kg min 70 max 110	Hint. Sitz Zuladung incl. Fallschirm in kg min 0 max 110	Alte Trudelballast-Tabelle* entfernt (abhaken) —	Signatur des Prüfers, Prüf-stempel  M. W. K.
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\* Flughandbuch Seite 10d

Änd.Nr. / Datum Sig.  
 TM 4a Nov. 04 JN/MG

Autor  
 Kaiser

Datum  
 April 80

Seite 11

LBA-amerk.

CAUTION : Incorrect loading can deteriorate glider handling qualities and can cause hazardous flight conditions. The pilot in command is responsible for correct loading.

Never fly the glider from the rear seat only !!

## VI.4 Calculation of CG at flight weight

	Weight [lbs]	x	arm [inch]	=	Momentum [lbs x inch]
Empty weight		x +		=	
Front pilot		x -	{ 46,65 } * { 49, 21 }	=	
Rear pilot		x -		=	
Baggage		x +		=	
<u>Sum of weight</u> →			<u>Sum of momentum</u> →		

$$\text{Position of flight CG} = \frac{\text{Sum of momentum}}{\text{Sum of weight}} = \boxed{\phantom{000}} \xrightarrow{\text{CG Flight [ inches]}}$$

\* Note: Tall persons shall use the shorter value and set the backrest on the rear position.  
Small persons shall use the longer value and set the backrest at the forward position.



## VI.4 Calculation of CG at flight weight [metric system]

	Weight [ kg ]	x	arm [ mm ]	=	Momentum [ kg x mm ]
Empty weight		x +		=	
Front pilot		x -	$\left\{ \begin{matrix} 1185 \\ 1250 \end{matrix} \right\}^*$	=	
Rear pilot		x -		=	
Baggage		x +		=	
<u>Sum of weight</u>					
			<u>Sum of momentum</u>		

Position of flight CG =  $\frac{\text{Sum of momentum}}{\text{Sum of weight}} = \boxed{\phantom{00000}}$  CG Flight [ mm ]

\* Note: Tall persons shall use the shorter value and set the backrest on the rear position.

Small persons shall use the longer value and set the backrest at the forward position.

## VI.4 Calculation of CG at flight weight

EXAMPLE !

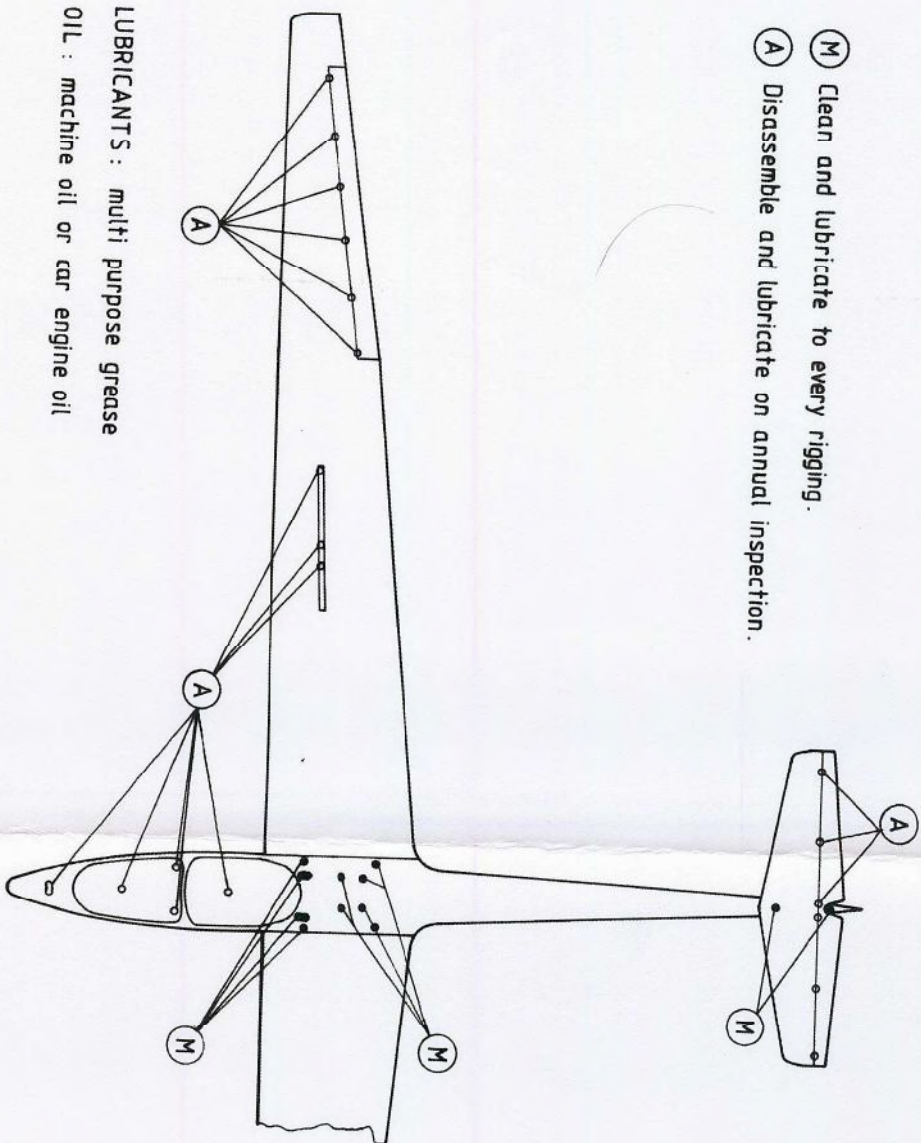
	Weight [lbs]	x	arm [inch]	=	Momentum [lbs x inch]
Empty weight	814	x +	30,55	=	+ 24 869
Front pilot	187	x -	$\left\{ \begin{array}{l} 46,65 \\ 49,21 \end{array} \right\}^*$	=	- 8833,88
Rear pilot	165	x -	3,15	=	- 519,75
Baggage	22	x +	9,84	=	+ 216,48
Sum of weight	1188		Sum of momentum		15730,85

$$\text{Position of flight CG} = \frac{\text{Sum of momentum}}{\text{Sum of weight}} = \boxed{13,24} \quad \text{CG Flight [ inches ]}$$

\* Note: Tall persons shall use the shorter value and set the backrest on the rear position.  
 Small persons shall use the longer value and set the backrest at the forward position.

## LUBRICATION SCHEME

- (M) Clean and lubricate to every rigging.
- (A) Disassemble and lubricate on annual inspection.



LUBRICANTS : multi purpose grease

OIL : machine oil or car engine oil



# Three view - Rigging data

